HEAT TRANSFER

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- Conduction problems may involve multiple directions and time-dependent conditions
- > Inherently complex-difficult to determine temperature distributions.
- In a one-dimensional system, temperature gradients exist along only a single coordinate direction, and heat transfer (diffusion) occurs exclusively in that direction.
- > In steady-state conditions, the temperature at each point is independent of time.
- > First, the heat transfer under this conditions will be discussed considering that there is no heat generation.
- > The heat resistance will be introduced.

Methodology of a conduction analysis

- 1. Specify appropriate form of the heat equation
- 2. Solve for the temperature distribution
- 3. Apply Fourier's law to determine the heat flux

Simplest case:

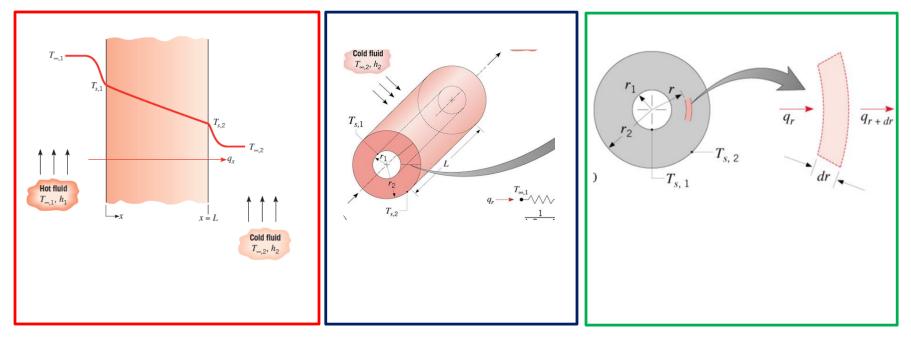
One-dimensional, steady state conduction with no thermal energy generation

Common geometries:

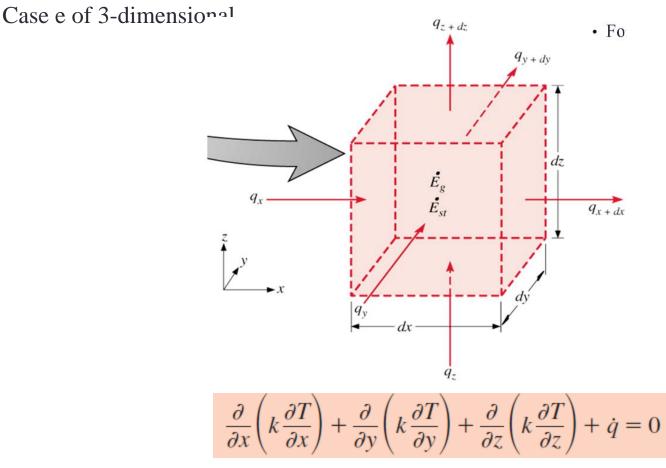
- The plane wall: described in rectangular (x) coordinate. Area perpendicular to direction of heat transfer is constant (independent of x).
- Cylindrical wall: radial conduction through tube wall.
- Spherical wall: radial conduction through shell wall.

Methodology of a conduction analysis

Common geometries



1. The Plane Wall



1. The Plane Wall

Using the following equation:

$$\frac{\partial}{\partial x} \left(k \frac{\partial T}{\partial x} \right) + \frac{\partial}{\partial y} \left(k \frac{\partial T}{\partial y} \right) + \frac{\partial}{\partial z} \left(k \frac{\partial T}{\partial z} \right) + \dot{q} = 0$$

Considering: (I) no heat generation, (II) steadystate conditions and (Ii) one-dimensional flow:

$$\frac{d}{dx}\left(k\frac{dT}{dx}\right) = 0$$

For constant k and A, this equation will be a second order differential equation:

$$\frac{d^2T}{dx^2} = 0$$

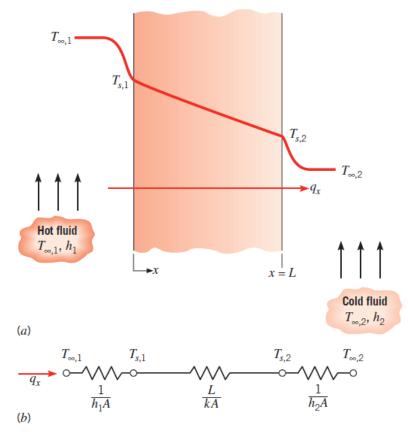


FIGURE 3.1 Heat transfer through a plane wall. (*a*) Temperature distribution. (*b*) Equivalent thermal circuit.

1. The Plane Wall

1-D heat conduction equation for steady-state conditions and no internal heat generation (i.e. q = 0), is

 $\frac{d^2T}{dx^2} = 0 \qquad \text{for constant } k \text{ and } A$

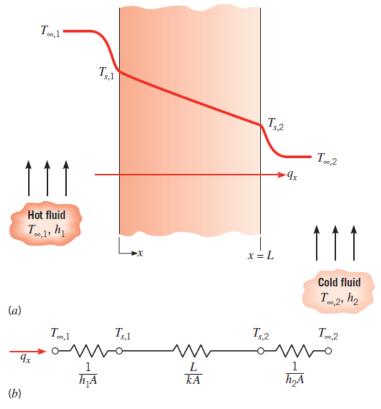
This mean: Heat flux (q''_x) is independent of x Heat rate (q_x) is independent of x

The general solution of the equation is:

$$T(x) = \iint \frac{d^2 T}{dx^2} dx dx = \int \left(\frac{dT}{dx} + C_1\right) dx$$

$$\therefore T(x) = C_1 \cdot x + C_2$$

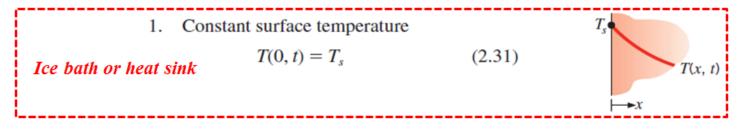
Where C1 and C1, are constants which can be found from the *boundary conditions* of the system.



1. The Plane Wall

Boundary conditions

In this system, first kind boundaries were used



For boundary conditions: $T(0) = T_{s,1}$ and $T(L) = T_{s,2}$

at
$$x = 0$$
, $T(x) = T_{s,1}$ and $C_2 = T_{s,1}$
at $x = L$, $T(x) = T_{s,2}$ and $T_{s,2} = C_1 L + C_2 = C_1 L + T_{s,1}$
this gives, $C_1 = (T_{s,2} - T_{s,1})/2$

Using value of C_1 and C_2 , the function of T(x) is

$$T(x) = (T_{s,2} - T_{s,1})\frac{x}{L} + T_{s,1} \qquad \Longrightarrow \quad \frac{dT}{dx} = \frac{T_{s,2} - T_{s,1}}{L}$$

<u>From here, apply Fourier's</u> law to get heat transfer.

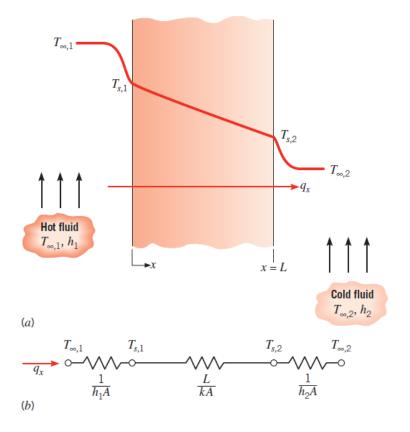
1. The Plane Wall

Heat flux for plane wall:

$$q_x'' = -k\frac{dT}{dx} = \frac{k}{L}\left(T_{s,1} - T_{s,2}\right)$$

Heat rate for plane wall:

$$q_x = -kA\frac{dT}{dx} = \frac{kA}{L} \left(T_{s,1} - T_{s,2}\right)$$



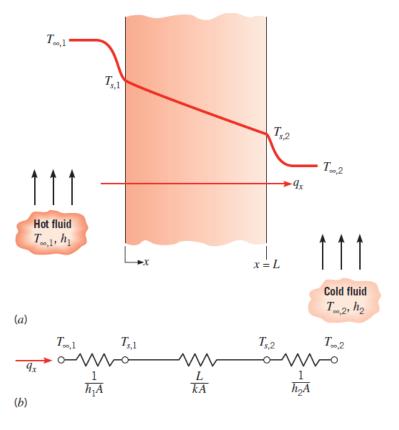
1. The Plane Wall

Heat flux for plane wall:

$$q_x'' = \frac{q_x}{A} = \frac{k}{L} (T_{s,1} - T_{s,2})$$

Heat rate for plane wall:

$$q_x = -kA \frac{dT}{dx} = \frac{kA}{L} \left(T_{s,1} - T_{s,2}\right)$$



2. Thermal Resistance

Just like the electrical conductivity and electrical resistivity, the thermal conductivity and thermal resistivity are related to each other:

Recall electric circuit theory - Ohm's law for electrical resistance:

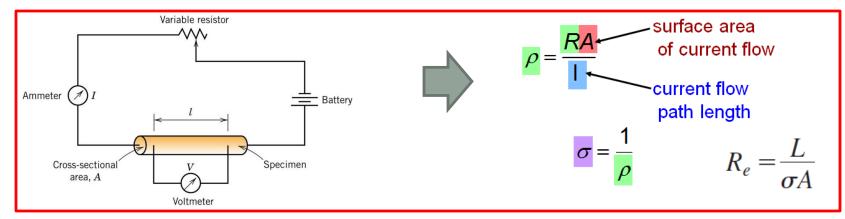
$$Electric \ current = \frac{Potential \ Differenc \ e}{Resistance}$$

> By the same way, the heat transfer rate equation can be written as:

$$-kA\frac{dT}{dx} = \frac{kA}{L}(T_{s,1} - T_{s,2}) = \frac{(T_{s,1} - T_{s,2})}{L/kA}$$

2. Thermal Resistance

Just like the electrical conductivity and electrical resistivity, the thermal conductivity and thermal resistivity are related to each other:



$$R_{t,\text{cond}} = \frac{L}{kA} \equiv \frac{T_{s,1} - T_{s,2}}{q_x}$$

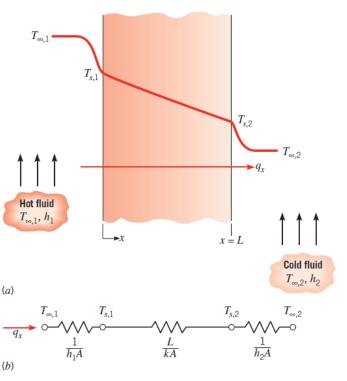
2. Thermal Resistance

Considering the convection heat transfer at the surface:

$$q = hA(T_s - T_{\infty})$$
 $R_{t,conv} \equiv \frac{T_s - T_{\infty}}{q} = \frac{1}{hA}$

- In this regard, the modelling of heat resistance using circuit representations will be needed to include both modes.
- The equivalent thermal circuit for the plane wall with convection surface conditions is (a) shown here:
- \triangleright Since q_x is constant throughout the system:

$$q_x = \frac{T_{\infty,1} - T_{s,1}}{1/h_1 A} = \frac{T_{s,1} - T_{s,2}}{L/kA} = \frac{T_{s,2} - T_{\infty,2}}{1/h_2 A}$$



2. Thermal Resistance

We can use this electrical analogy to represent heat transfer problems using the concept of *a thermal circuit* (equivalent to an electrical circuit).

$$q_{x} = \frac{\text{Overall Driving Force}}{\text{Resistance}} = \frac{\Delta T_{overall}}{\sum R}$$
$$q_{x} = \frac{T_{\infty,1} - T_{s,1}}{1/h_{1}A} = \frac{T_{s,1} - T_{s,2}}{L/kA} = \frac{T_{s,2} - T_{\infty,2}}{1/h_{2}A} \implies q_{x} = \frac{T_{\infty,1} - T_{\infty,2}}{R_{\text{tot}}}$$

The conduction and convection resistances are in series and can be summed

$$\begin{array}{c} T_{\infty,1} & T_{s,1} & T_{s,2} & T_{\infty,2} \\ \hline q_x & & & \\ \hline \frac{1}{h_1A} & & & \\ \hline \frac{L}{kA} & & & \\ \hline \frac{1}{h_2A} & & \\ \end{array} \end{array}$$

$$R_{\text{tot}} = \frac{1}{h_1A} + \frac{L}{kA} + \frac{1}{h_2A}$$

3. The Composite Wall

The thermal circuits for more complex systems, such as composite walls, in which various layers can be included in the wall.

For the wall shown here, the following equation can be used to introduce the heat transfer rate:

$$q_x = \frac{T_{\infty,1} - T_{\infty,4}}{\Sigma R_t}$$

where $T_{\infty,1}$ - $T_{\infty,4}$ is the overall temperature difference.

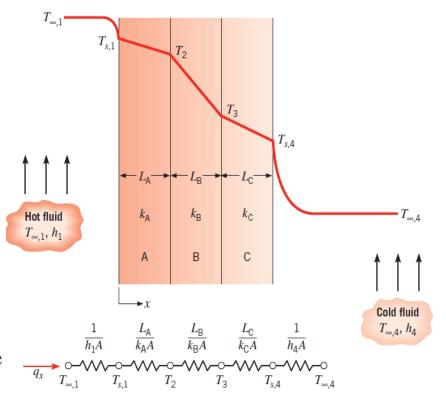


FIGURE 3.2 Equivalent thermal circuit for a series composite wall.

One-Dimensional, Steady-State Conduction 3. The Composite Wall

$$q_x = \frac{T_{\infty,1} - T_{\infty,4}}{\left[(1/h_1A) + (L_A/k_AA) + (L_B/k_BA) + (L_C/k_CA) + (1/h_4A)\right]}$$

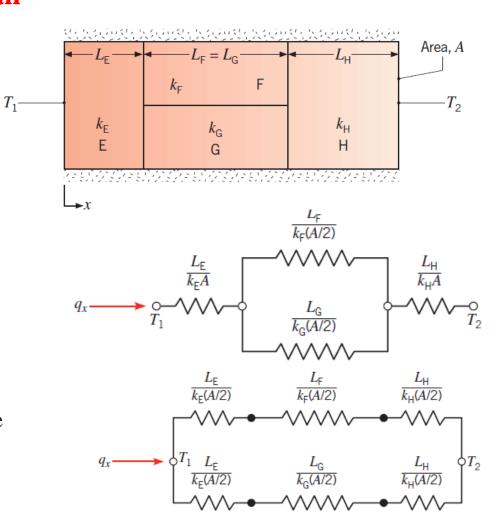
For this wall, the overall heat transfer coefficient U is used:

$$U = \frac{1}{R_{\text{tot}}A} = \frac{1}{[(1/h_1) + (L_A/k_A) + (L_B/k_B) + (L_C/k_C) + (1/h_4)]}$$
$$Q_x \equiv UA \ \Delta T$$
$$R_{\text{tot}} = \sum R_t = \frac{\Delta T}{q} = \frac{1}{UA}$$

3. The Composite Wall Series-parallel type

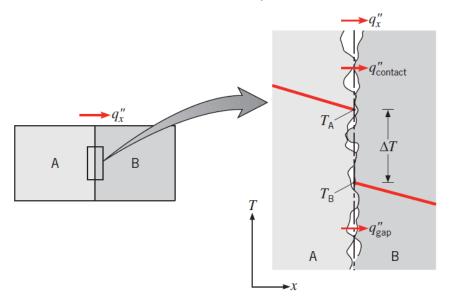
1. Surfaces normal to the *x*-direction are isothermal

2. Surfaces parallel to the x-direction are adiabatic



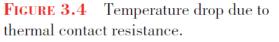
4. Contact Resistance

- Although neglected until now, it is important to recognize that, in composite systems, the temperature drop across the interface between materials may be appreciable.
- > This temperature change is attributed to what is known as the thermal contact resistance, $R_{t,c}$.



$$R_{t,c}'' = \frac{T_{\rm A} - T_{\rm B}}{q_x''}$$

$$q''_X = q''_{gap} + q''_{cont}$$



4. Contact Resistance

TABLE 3.1 Thermal contact resistance for (a) metallic interfaces under vacuum conditions and (b) aluminum interface (10-μm surface roughness, 10⁵ N/m²) with different interfacial fluids [1]

Thermal Resistance, $R''_{t,c} \times 10^4 \,(\text{m}^2 \cdot \text{K/W})$

(a) Vacuum Interface			(b) Interfacial Fluid		
Contact pressure	100 kN/m ²	10,000 kN/m ²	Air	2.75	
Stainless steel	6-25	0.7-4.0	Helium	1.05	
Copper	1–10	0.1-0.5	Hydrogen	0.720	
Magnesium	1.5-3.5	0.2-0.4	Silicone oil	0.525	
Aluminum	1.5-5.0	0.2-0.4	Glycerine	0.265	

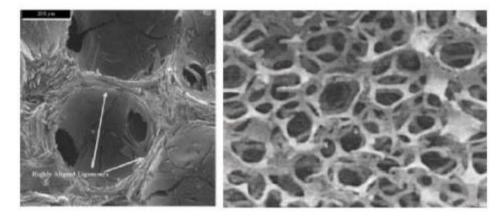
4. Contact Resistance

TABLE 3.2	Thermal resistance	of represe	ntative solid	/solid	interfaces
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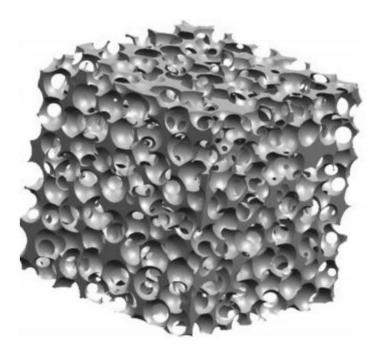
Interface	$R_{t,c}^{\prime\prime} \times 10^4 (\mathrm{m}^2 \cdot \mathrm{K/W})$	Source
Silicon chip/lapped aluminum in air (27–500 kN/m ²)	0.3–0.6	[2]
Aluminum/aluminum with indium foil filler (~100 kN/m ²)	~0.07	[1, 3]
Stainless/stainless with indium foil filler (\sim 3500 kN/m ²)	~ 0.04	[1, 3]
Aluminum/aluminum with metallic (Pb) coating	0.01-0.1	[4]
Aluminum/aluminum with Dow Corning 340 grease ($\sim 100 \text{ kN/m}^2$)	~0.07	[1, 3]
Stainless/stainless with Dow Corning 340 grease (~3500 kN/m ²)	$\sim \! 0.04$	[1, 3]
Silicon chip/aluminum with 0.02-mm epoxy	0.2-0.9	[5]
Brass/brass with 15- μ m tin solder	0.025-0.14	[6]

5. Porous Media

In many applications, heat transfer occurs within *porous media* that are combinations of a stationary solid and a fluid.



Photographs of (a) Scanning Electron Microscope (SEM) image showing the cross section of carbon foam, and (b) an aluminum foam consisting of interconnected ligaments



5. Porous Media

Consider a saturated porous medium that is subjected to surface temperatures T_1 at x = 0 and T_2 at x = L. After steady-state conditions are reached and if T_1 T_2 , the heat rate may be expressed as:

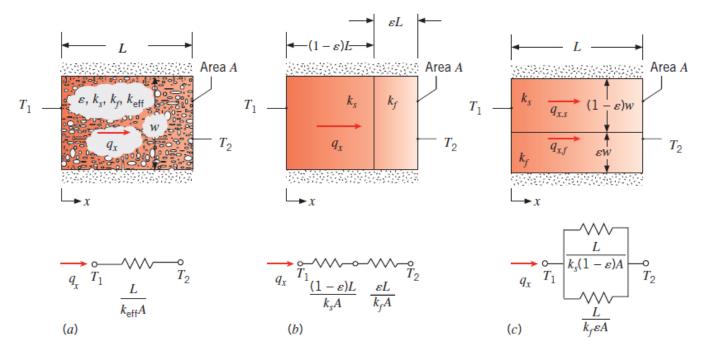


FIGURE 3.5 A porous medium. (*a*) The medium and its properties. (*b*) Series thermal resistance representation. (*c*) Parallel resistance representation.

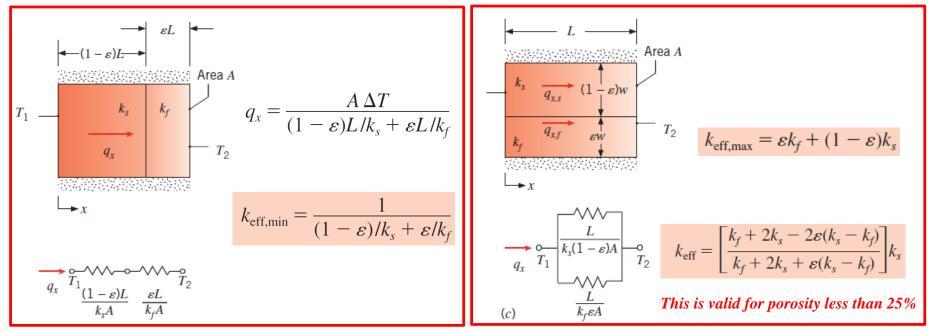
5. Porous Media

Heat rate may be expressed as

$$q_x = \frac{k_{\rm eff}A}{L} \left(T_1 - T_2\right)$$

where k_{eff} is an effective thermal conductivity of the porous system.

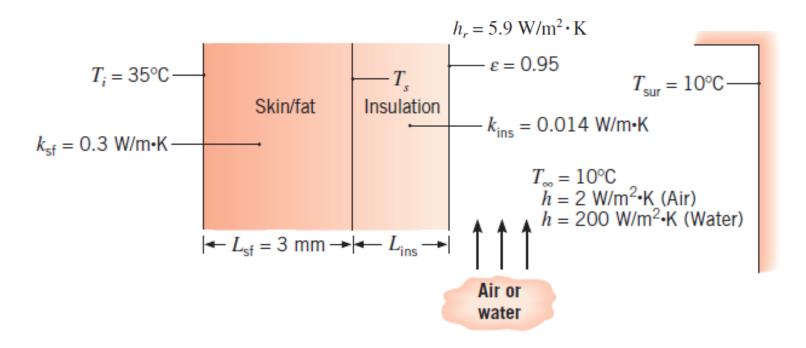
This equation is valid when there is no radiation heat transfer.



1. The Plane Wall

Example-1

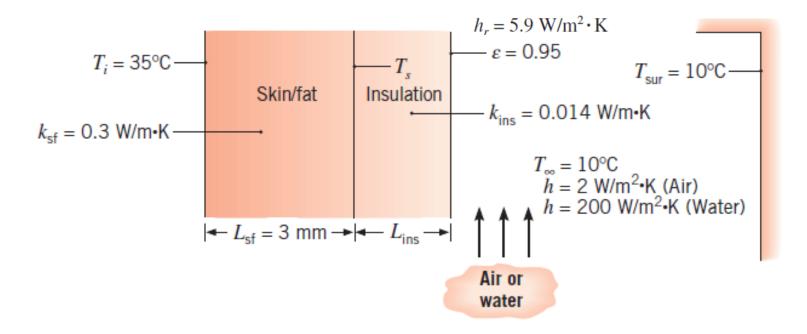
- → Heat loss from the skin surface is (I) 146 W (air) and (II) 1320 W (water).
- Calculate the thickness of the insulator needed to reduce the heat transfer rate from this skin to 100 W in both cases (air and water).



1. The Plane Wall

Example-1

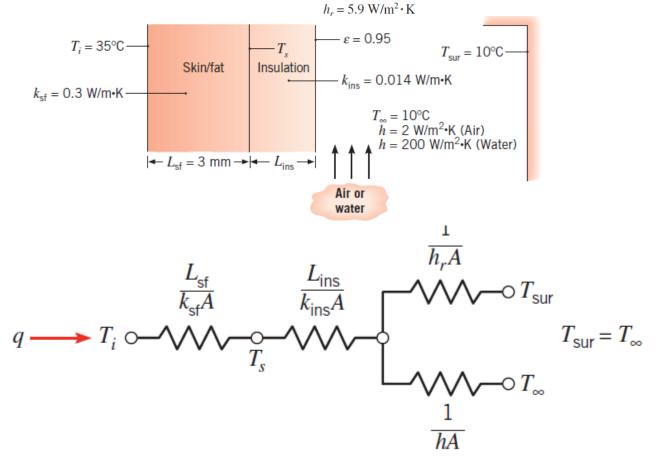
- ➢ To solve this problem, first the heat transfer modes, by which the heat from the skin is lost, should be introduced.
- > Total resistance concept should be used.



1. The Plane Wall Example-1

- > Steady-state conditions.
- One-dimensional heat transfer by conduction through the skin/fat and insulation layers.
- > Contact resistance is negligible.
- Radiation exchange between the skin surface and the surroundings is between a small surface and a large enclosure at the air temperature.
- > Liquid water is opaque to thermal radiation.

1. The Plane Wall Example-1



1. The Plane Wall Example-1

$$R_{\text{tot}} = \frac{T_i - T_{\infty}}{q} = \frac{(35 - 10) \text{ K}}{100 \text{ W}} = 0.25 \text{ K/W}$$

$$R_{\text{tot}} = \frac{L_{\text{sf}}}{k_{\text{sf}}A} + \frac{L_{\text{ins}}}{k_{\text{ins}}A} + \left(\frac{1}{1/hA} + \frac{1}{1/h_rA}\right)^{-1} = \frac{1}{A} \left(\frac{L_{\text{sf}}}{k_{\text{sf}}} + \frac{L_{\text{ins}}}{k_{\text{ins}}} + \frac{1}{h + h_r}\right)$$

$$\frac{Air}{L_{\text{ins}}} = k_{\text{ins}} \left[AR_{\text{tot}} - \frac{L_{\text{sf}}}{k_{\text{sf}}} - \frac{1}{h + h_r}\right]$$

$$= 0.014 \text{ W/m} \cdot \text{K} \left[1.8 \text{ m}^2 \times 0.25 \text{ K/W} - \frac{3 \times 10^{-3} \text{ m}}{0.3 \text{ W/m} \cdot \text{K}} - \frac{1}{(2 + 5.9) \text{ W/m}^2 \cdot \text{K}}\right]$$

$$= 0.0044 \text{ m} = 4.4 \text{ mm}$$

1. The Plane Wall Example-1

<u>Water</u>

$$R_{\text{tot}} = \frac{L_{\text{sf}}}{k_{\text{sf}}A} + \frac{L_{\text{ins}}}{k_{\text{ins}}A} + \left(\frac{1}{1/hA} + \frac{1}{1/h_rA}\right)^{-1} = \frac{1}{A} \left(\frac{L_{\text{sf}}}{k_{\text{sf}}} + \frac{L_{\text{ins}}}{k_{\text{ins}}} + \frac{1}{h + h_r}\right)^{-1}$$
$$L_{\text{ins}} = k_{\text{ins}} \left[AR_{\text{tot}} - \frac{L_{\text{sf}}}{k_{\text{sf}}} - \frac{1}{h}\right]$$
$$= 0.014 \text{ W/m} \cdot \text{K} \left[1.8 \text{ m}^2 \times 0.25 \text{ K/W} - \frac{3 \times 10^{-3} \text{ m}}{0.3 \text{ W/m} \cdot \text{K}} - \frac{1}{200 \text{ W/m}^2 \cdot \text{K}}\right]$$
$$= 0.0061 \text{ m} = 6.1 \text{ mm}$$

1. The Plane Wall Example-1

The skin surface temperature can be calculated by considering conduction through the skin/fat layer:

